AMENDMENTS TO THE CLAIMS:

This listing of claims will replace all prior versions, and listings, of claims in this application.

1. (Original) An acoustic panel for use in the inlet lip portion of a gas turbine nacelle, the panel comprising: (a) a solid back skin; (b) an acoustically permeable front skin; (c) a honeycomb cell structure located between the front skin and back skin; and (d) an ice protection system affixed to the front skin, wherein the ice protection system includes an acoustically permeable and electrically and thermally conductive structure which includes means for connection to an electric power source, and the structure is thermally insulated from the front skin.

2. (Original) The acoustic panel of Claim 1, in which the ice protection system includes a low power electronic ice protection system.

3. (Original) The acoustic panel of Claim 1, in which the acoustically permeable front skin is perforated.

4. (Original) The acoustic panel of Claim 1, in which the honeycomb structure is adhesively bonded to the front skin and the back skin.

5. (Original) The acoustic panel of Claim 1, in which the front skin and the back skin are each an aluminum sheet material.

6. (Original) The acoustic panel of Claim 1, in which the front skin, back skin and honeycomb cell structure are each a graphite/epoxy laminate.

7. (Original) The acoustic panel of Claim 1, in which the ice protection system includes a stainless steel wire mesh adhesively bonded to the outer surface of the front skin.

8. (Original) The acoustic panel of Claim 1, in which a permeable, thermally insulating material is located between the electronic ice protection system and the front skin.

9. (Original) The acoustic panel of Claim 8, in which the insulating material is adhesively bonded to the outer surface of the front skin, and the electronic ice protection system is adhesively bonded to the insulating material.

10. (Previously presented) The acoustic panel of Claim 1, in which the nacelle has a highlight, and a parting strip is located proximate to the nacelle highlight.

11. (Original) The acoustic panel of Claim 10, in which the parting strip is an electrified grid material which carries a watt density of up to about 20W/sq. in.

12. (Previously presented) The acoustic panel of Claim 1, in which the ice protection system comprises a plurality of sections which extend around the circumference of the inlet lip of the nacelle.

13. (Original) The acoustic panel of Claim 12, in which power is supplied selectively or sequentially to the sections.

14. (Cancelled)

15. (Cancelled)

16. (Original) An inlet lip for an aircraft gas turbine engine nacelle, the inlet lip comprising: (a) an acoustic panel structure including a solid back skin, an acoustically

permeable front skin, and a honeycomb cell structure there between; and (b) an ice protection system located on the front skin, wherein the ice protection system includes an acoustically permeable and electrically and thermally conductive structure in electrical connection to an electric power source, and the ice protection system is thermally insulated from the permeable front skin.

17. (Cancelled)

18. (Cancelled)

19. (Cancelled)

20. (Cancelled)